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measurements using  
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AIMMS20AQ**

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# Application of the Aventech AIMMS20AQ airborne probe for turbulence measurements during the Convective Storm Initiation Project

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## Abstract

The Convective Storm Initiation Project (CSIP) took place during the summers of 2004 and 2005, centred on the research radar at Chilbolton, UK. Precursors to convective precipitation were studied, using a comprehensive and broad-based range of fieldwork and modelling. The principal aim of CSIP was the detection of the primary and secondary initiation of convective cells. The Universities Facility for Atmospheric Measurements (UFAM) Cessna 182 was used to map temperature and humidity fields over a broad area within and beyond the Chilbolton radar beam. Additionally, air motion was measured using a new turbulence probe, the AIMMS20AQ. The performance of the probe is critically appraised, based on calibrations, test flights and data flights flown during CSIP intensive operating periods. In general, the probe performed well, although some aspects require more careful data interpretation which we describe in detail.

## 1 Introduction

With the expected increase in the frequency, magnitude and effects of extreme weather events due to global warming (IPCC, 2001; Senior et al., 2002; McBean, 2004), it has become crucial for governments to be able to plan for both the short and long-term consequences of these events. In particular, there is a pressing need to issue suitable and timely warnings where imminent danger to population and infrastructure is expected.

The Convective Storm Initiation Project (CSIP) was set up to improve the spatial and temporal accuracy of forecasting of high precipitation events. Since the accuracy of existing forecasts was greatly dependent on the success of the forecast of the initial storm development, CSIP targeted the main mechanisms thought to be precursors of convective storms. Prior to CSIP, the scale of the measurements required to provide adequate data to initialise and verify forecasting models meant there had been no field

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studies focussing on the initiation of precipitating convection in the UK. The recently established NERC Centre for Atmospheric Sciences UFAM facility (Universities Facility for Atmospheric Measurement) allowed a comprehensive, broad-based approach, covering all stages of storm development, including cases where convection was expected but failed to produce precipitation.

We report in this paper airborne measurements of wind and turbulence structure using the UFAM Cessna 182J during the CSIP programme. The measurements were made using a new fast-response probe, the AIMMS20AQ (manufactured by Aventech of Barrie, Ontario). We present two contrasting case studies of wind fields, and include an assessment of the performance of the instrument based on comparisons with a ground-based wind profiler and network of automatic weather stations.

## 2 CSIP methodology

Fieldwork for CSIP consisted of a pilot project in June 2004, followed by a full intensive campaign from June to August 2005, with the pilot project being used to inform and optimise the use of equipment during the main campaign.

The work was centred around the Council for the Central Laboratory of the Research Councils (CCLRC) site at Chilbolton, on the eastern edge of Salisbury Plain (Goddard et al., 1994; Naud et al., 2005). The two primary instruments mounted on the 25 m radar dish at this site are the Chilbolton Advanced Meteorological Radar Antenna (CAMRA), which can identify boundary-layer structures such as thermals and cloud shapes, and also give wind and moisture fields, and ACROBAT – the Advanced Clear Air Radar for Observing the Boundary Layer and Troposphere. For CSIP, the radar measurements were supplemented with a UHF wind profiler, sodars and Doppler Lidars stationed at sites within the radar range, as well as a mesonet of automatic weather stations spaced at roughly 20 km intervals. Radiosondes were launched at up to hourly intervals from six sites. The UFAM Cessna and the Institut für Meteorologie und Klimaforschung (IMK) Dornier 128 were flown during selected conditions. CSIP

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data from these instruments will be presented elsewhere, and an overview of the CSIP project is given by Browning et al. (2006) who also give details of the forecasting and modelling products used.

On any given day, the probability of the onset of convective precipitation was assessed by the use of the model early in the day, with a forecast available by 09:00 GMT. If suitable conditions were forecast, then the day would be declared an Intensive Operating Period (IOP). The forecast was then used to determine the optimum flight plan for the Cessna and Dornier aircraft.

Three pre-determined flight paths were used, based on the set of navigational “waypoints” shown in Fig. 1, giving a consistent dataset which would enable direct comparisons between flights. The first route was a simple East-West transit across Salisbury Plain between Greenham Common (waypoint J) and Bath (O). The second was the “Northern Box”, extending the East-West transit to the North using different combinations of waypoints depending on conditions. The third flight path was the “Southern Box” with a loop down to the south coast before heading up to Bath, mainly flown by the IMK Dornier. A fourth path was flown on the last day of the main project, with a North-South transit between Chilbolton and Swindon flown several times. Ideally, all flight paths were to be flown at a constant altitude of 2000 feet (610 m), with adjustments for individual flights made according to air traffic control instructions or local weather conditions.

A total of 18 IOP days were declared for the full CSIP campaign in 2005. The Cessna flew single flights on four of these days and two flights on three others: Table 1 lists the flights, indicating the routes flown.

### 3 Cessna instrumentation

The UFAM Cessna 182 J is a single-engine, high-winged aircraft, capable of flying at an altitude up to 3000 m, with an endurance of around 4 h. The normal cruising speed is 110 knots (approximately  $55 \text{ ms}^{-1}$ ) with a crew of one pilot and one observer, and

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5 a payload of up to 130 kg. The aircraft has been owned and run by the University of Manchester since 1984 and has been used on a wide variety of research work e.g. Smith et al. (1989), Gallagher et al. (1990, 1994), Wood et al. (1999), Bower et al. (2000) and Webb et al. (2004). Although normally hangared at Liverpool John Lennon Airport, for the duration of the CSIP campaign the aircraft was kept at Thruxton Airfield, in the centre of the CSIP operational area allowing a quick response to changing conditions.

The principal measurements for CSIP were position, temperature, humidity and air motion. The Cessna has a range of standard instruments for pressure, air speed and navigation, and these are supplemented by a suite of scientific instruments.

10 Air temperature was measured by three separate probes. The primary instrument was a constant current platinum wire total temperature probe in a wing-mounted Rosemount inlet. The response time for this sensor was approximately 1 s (Wood, 1997). A second temperature sensor was mounted in a reverse-flow housing, allowing it to operate in wet and dry conditions. A second reverse-flow thermometer housed within the AIMMS turbulence probe is detailed in Sect. 5. All temperature data were corrected for dynamic heating effects (Lenschow, 1986) using recovery factors determined specifically for the Cessna instruments (Wood et al., 1997).

20 Absolute and relative humidity were derived from the Rosemount temperature and the dewpoint signal from a Michell S3000 Hygrometer. The response time as determined from spectral analysis of field data was 5 s (Wood, 1997). The AIMMS relative humidity sensor is discussed in Sect. 5 (response 1 Hz, Private communication, Bruce Woodcock, Aventech).

25 A Javad Navigation Systems AT4 differential GPS (DGPS) system was used to determine position at 10 Hz (the current equivalent model is the JNSGyro-4T). The AT4 uses four geodetic quality antennae arranged in a cruciform on the tail, wings and cabin roof, with a baseline of 5.3 m between the wing antennae. The antennae are capable of tracking up to 20 satellites each, and use the dual frequency (1565–1615 MHz and 1217–1265 MHz) carrier-phase method to greatly enhance the accuracy of the mea-

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surements. As well as providing accurate position, by calibrating the antennae on the ground over a period of around an hour, the positions of the antennae are known to an accuracy of around 1 mm relative to each other in three dimensions. By defining the tail antenna as the "master" and the other three antennae as "slaves", the AT4 can then calculate aircraft attitude angles (pitch, roll and heading) to an accuracy of 0.1°, and position in three dimensions to within 0.01 m. This system is also used on the UFAM BAe 146 aircraft. For the Cessna application, the AT4 uses the standard NMEA \$GPRMC command (DePriest, worldwide web) to output position, ground track and ground speed, as well as satellite time. Additionally, the three attitude angles are output via a command unique to the AT4.

The two wing antennae for the AT4 are shared with the GPS module which forms part of the AIMMS probe (see Sect. 5). Whilst altitude is available from the two GPS systems, this only gives height above the GPS geoid, not above sea level. Thus, for the purposes of this paper, altitude is calculated from the aircraft static pressure signal.

All signals were logged to a custom-built PC-based logging system, comprising two PCs running Windows XP, a 32 channel differential A-to-D interface and an Amplicon 8-port serial card. The standard aircraft instruments and the AT4 were logged using custom-written LabView programmes, whilst the AIMMS was logged using software supplied with the probe.

## 4 Air velocity measurements from aircraft platforms

### 4.1 Theory

The true wind velocity  $V$  of atmospheric air relative to the surface of the Earth is found from the following vector sum:

$$V = V_a + V_p \quad (1)$$

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where  $\mathbf{V}_a$  is the air velocity relative to the moving aircraft, and  $\mathbf{V}_p$  is the velocity of the aircraft with respect to the Earth. Thus  $\mathbf{V}$  is the difference between two large quantities and it is therefore essential that  $\mathbf{V}_a$  and  $\mathbf{V}_p$  are measured as accurately as possible in order to give an acceptable level of error in  $\mathbf{V}$ , usually of the order of 0.5–1.0 ms<sup>-1</sup> for meteorological research. Additionally, in order to be able to study small-scale structures, the frequency at which these quantities are measured must be suitable relative to the speed of the aircraft.

Equation (1) requires  $\mathbf{V}_a$  and  $\mathbf{V}_p$  to be expressed in the same coordinate system.  $\mathbf{V}_p$  is defined relative to the Earth by three attitude angles, pitch  $\theta$ , roll  $\phi$  and heading  $\psi$ , as shown in Fig. 2. The Earth coordinate axes are  $x$ ,  $y$  and  $z$ , (collectively known as  $S$ ) where  $x$  is positive in the East direction,  $y$  is positive in the North direction, and  $z$  is positive in the upwards vertical direction.  $\mathbf{V}_a$  is measured relative to the aircraft in the  $x'y'z'$  coordinate system (collectively known as  $S'$ ), where  $x'$  is the longitudinal axis of the aircraft,  $y'$  is the lateral axis, and  $z'$  is the vertical axis, and is defined by two angles: vertical angle of attack,  $\alpha$ ; horizontal sideslip angle,  $\beta$ .  $\mathbf{V}_a$  must therefore be mapped from  $S'$  to  $S$  to allow Eq. (1) to be calculated.

Figure 2 shows the relationship between the  $S$  and  $S'$  coordinate systems. A matrix,  $\mathbf{M}_{S' \rightarrow S}$ , is used to map  $S'$  to  $S$ :

$$\mathbf{M}_{S' \rightarrow S} = \begin{pmatrix} \cos \theta \sin \Psi & -\sin \phi \sin \theta \sin \Psi - \cos \phi \cos \Psi & -\cos \phi \sin \theta \sin \Psi + \sin \phi \cos \Psi \\ \cos \theta \cos \Psi & -\sin \phi \sin \theta \cos \Psi + \cos \phi \sin \Psi & -\cos \phi \sin \theta \cos \Psi - \sin \phi \sin \Psi \\ \sin \theta & \sin \phi \cos \theta & \cos \phi \cos \theta \end{pmatrix} \quad (2)$$

The three rows of  $\mathbf{M}_{S' \rightarrow S}$  transform the longitudinal, lateral and vertical components of  $\mathbf{V}_a$ . The vector components of  $\mathbf{V}_a$  and  $\mathbf{V}_p$  can then be used in Eq. (1) to give the air motion vector in terms of north, east and vertical components. A full derivation of the transform matrix shown in Eq. (2) can be found in Wood (1997).

## 4.2 Turbulence probes

The most obvious difficulty in measuring air turbulence from any moving object is the removal of the platform velocity vector. Measuring turbulence from a fast-moving aircraft platform adds the further problem of removing significant airflow distortion induced by the aircraft itself. Combined with the need to measure the data as fast as possible in order to improve spatial resolution, it is only recently that technology has been able to overcome these problems satisfactorily.

A number of systems have been developed over the years. Although successful measurements are possible using a simple 3-axis accelerometer system to determine the response of the aircraft to gusts of wind (e.g. Stromberg et al., 1989), this can be subject to large errors due to the mass of the aircraft damping the response to turbulence. Notess et al. (1954) coupled accelerometers with fixed wind vanes and a pitot-static aircraft true air speed (TAS) sensor, whilst Telford and Warner (1962) improved on this system with the introduction of the gyro-stabilised platform, still a major component of modern inertial navigation systems. Whilst fixed and rotating vanes (Johnson et al., 1978; Lenschow, 1971) have been used to measure airflow relative to the aircraft, it is now more common to use a differential pressure method e.g. Brown et al. (1983), this being more reliable particularly under icing conditions. Typically, a “five-hole” arrangement of pressure ports is used, such as that in the Rosemount AJ858 gust probe (e.g. Bange et al., 2002). A more advanced instrument is the BAT Probe (Hacker and Crawford, 1999) using a nine hole pressure port formation in conjunction with differential GPS and fast-response accelerometers to give wind vector measurements at up to 50 Hz. Whiteway et al. (2003) used this probe to study breaking gravity waves at the top of the tropopause.

A five-hole turbulence probe was developed for use on the instrumented Cessna 182J (Wood et al. 1997), and the AIMMS turbulence probe, described in detail below, is based on the same principles, but incorporates state-of-the-art technology including differential GPS and miniaturised components, and makes full use of the updated

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computing power to allow the use of the Kalman Filter signal processing method.

## 5 The AIMMS20AQ probe

### 5.1 Technical description

The AIMMS20AQ probe – Aircraft Integrated Meteorological Measuring System 20Hz – was developed from wind sensors used in a crop spraying application designed to help farmers meet regulations on pesticide application in the United States. In consequence, economies of scale ensure the probe is robust, reliable, relatively low cost and easy to install and calibrate. Whilst the Cessna uses the standard probe casing, the sensor head can also be supplied to fit a PMS pod to make it simple to fit on research aircraft which are already modified for this widely used type of canister.

The AIMMS is essentially an up-to-date five hole probe, with all elements of the sensor, data processing and analysis in a stand-alone package. The system consists of four modules which will be described further below: ADP – the air data probe, comprising a five-hole pressure port head, and built-in temperature and humidity sensors; GPS – global positioning system linked to antennae on each wing; IMU – inertial measurement unit; CPM – central processing module. The modules are linked by a high-speed digital serial link known as the CAN – Controller Area Network – which also carries power between the modules. Figure 3 shows a schematic of the modules, and the technical specifications are given in Table 2.

The ADP has a cruciform array of five pitot-static pressure ports situated on a hemispherical head which sits at the front end of a cylinder which also contains a ring of static pressure ports for determining aircraft TAS. The horizontal and vertical pairs detect sideslip angle and angle of attack respectively. The rear of the cylinder contains a reverse-flow housing for the temperature and humidity sensors. The cylinder is mounted below an aerofoil-shaped pylon which contains the transducer electronics. The ADP can be mounted onto a wing strut or, in the case of the Cessna, directly to

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the underside of the wing. The combination of the pylon and cylinder structure ensures that the pressure sensor sits upwind of, and below, the front edge of the wing, away from the influence of much of the induced airflow distortion. Weighing 3.6 kg and with a drag of 4.6 N at  $50 \text{ ms}^{-1}$ , the probe has little effect on aircraft performance. Combined with using fixing points on the Cessna wing originally used for a PMS canister, this simplified the installation of the new probe. The PMS (Particle Measuring Systems) canister has been a widely used method for mounting research instruments on aircraft for many years, making the modification process much simpler, both practically and in terms of meeting UK CAA and the European EASA regulations (CAA certificate number 9/218/M/5438 for this application).

The remaining three modules are mounted within the aircraft. As with the AT4 DGPS, the AIMMS GPS uses the carrier-phase method to give accuracy of  $0.1^\circ$  in the heading calculation. Only two antennae are required, and the wing antennae used for the AT4 are shared via signal splitters (GPS Networking Inc., Model LDCBS1X2).

Ideally, the IMU should be mounted as close to the centre of mass (CoM) of the aircraft as possible, and preferably on the aircraft's longitudinal axis. This ensures that the accelerometer signals are representative of the movement of the entire aircraft and are not due to rotation about the CoM or excessive flexing of the airframe. In the Cessna, the IMU is mounted on the longitudinal axis, but is located 3 m aft of the CoM. The IMU consists of a 3-axis accelerometer unit and a 3-axis system of rate gyroscopes, both running at 40 Hz. Combining the signals from the IMU and GPS, the full attitude and motion of the aircraft can be determined at a rate of up to 20 Hz.

The CPM comprises a Motorola DSP56F807 processor with a 16Mbit internal flash memory. Using the CAN, data is taken from the ADP, IMU and GPS modules and processed. There are two mutually exclusive data recording methods. The first – raw data capture – records all data that passes along the CAN to the logging PC's hard drive. This binary data file is post-processed using calibration coefficients (see Sect. 5.2), giving the full 20 Hz capability of the probe, with the output given in CSV ASCII format. This recording method allows the probe to be flown before all the calibration flights

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have been carried out. The second method uses the calibration coefficients to give on-line wind speed data at up to 5 Hz: this relies on the magnetic and airflow calibrations being available in order to give a reliable solution, since the data cannot be further processed. The data is stored on the CPM flash memory and then extracted at the end of a flight as a CSV ASCII file. In both cases, an Extended Kalman Filter is used (Kalman, 1960; Welch and Bishop, 2004), in which a running average of an estimate of the error in the wind solution is used to prevent the integrated accelerometer signal from drifting. The Kalman filter requires a “spin-up” period of around 2 s in order to operate correctly, requiring the signals from the IMU and GPS to be buffered in order to give an accurate wind vector solution.

The CPM communicates with an external computer via two RS232 ports at 156 kbaud. A configuration and logging program is supplied with the probe which allows the user to input calibration coefficients, decide which of the two serial ports is to be used, and to set the logging method to either on-line or raw data capture.

## 5.2 Calibration procedures

The AIMMS probe requires three separate calibrations, namely magnetic, aerodynamic and cross-axis. The on-line processing method requires that the magnetic and aerodynamic calibrations have been carried out, although reliable data is then restricted to straight and level flight. For fully processed data, giving accurate wind speeds in most conditions, all three calibrations are essential. For all three calibrations, the data are analysed by Aventech who then provide calibration coefficients.

The magnetic calibration is carried out on the ground. The aircraft is pointed to magnetic north and then makes a 360° turn at a steady rate over a period of two minutes. The resulting calibration coefficients are then programmed into the CPM.

The aerodynamic calibration is designed to quantify the effect of distortion of the airflow around the aircraft when flying in a straight line, and must be carried out in the air, in conditions of uniform and low wind speed. Two pilot-induced manoeuvres are carried out, shown schematically in Fig. 4a. Firstly the aircraft is “yawed” 10°, initially to

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the port side and then to the starboard side. This entails changing the aircraft heading whilst keeping the wings level, a procedure known as a flat turn. This is repeated at two different speeds (close to stall speed, and highest cruising speed) and in two reciprocal directions. This allows the probe's response to sideslip angle to be calculated, whilst removing the effect of the prevailing wind conditions. In the second manoeuvre, the aircraft is made to climb 2000 feet rapidly at constant speed before making a 180° turn and descending rapidly by 2000 feet. This allows the response to angle of attack to be quantified, again removing the effect of the prevailing wind.

The cross-axis calibration is again carried out in flight during calm and consistent conditions, and quantifies the response of the probe when the aircraft makes turns. A square box is flown in clockwise and anti-clockwise directions, with banked turns of different roll angle at each corner, as shown schematically in Fig. 4b.

### 5.3 Quality control

At the time of the CSIP campaign for which these results are reported, the cross-axis calibration had not been completed for the AIMMS20, and the aerodynamic calibration was not completed until after Flight 5. As a result, only data recorded during straight and level runs have been used. All data for flights 1 to 5 were made using the raw data mode but, for operational reasons, it was decided to use the on-line mode at 5Hz for subsequent flights, although full spectral analysis of the turbulence data would not be possible.

There were short periods where the AIMMS wind vector solution failed, in particular on the EF leg of Flight 7. The AT4 data, whilst generally of very high quality, had occasional “drop-outs” of a few seconds where the number of satellites visible to the antennae were insufficient to give reliable data: these periods have been removed.

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## 5.4 Data processing

In terms of importance for the CSIP campaign, the most useful output was for temperature, humidity and wind vector as a function of position along a given flight track. For the purposes of this paper, only the wind field data will be discussed in detail, where the data from the main loggers were averaged to 30 seconds. Additionally, in order to characterise the abilities of the probe, the original on-line processed 5 Hz data were used for the AIMMS probe.

The AIMMS was logged to one of the on-board PCs, with the ASCII data processed through the proprietary programmes Microsoft Excel and Wavemetrics Igor Pro. The second PC was used to log all other data: the Javad AT4 was logged at 10 Hz via one of the standard serial ports; the remaining signals were logged via a 32 channel A/D card. The A/D was logged at 20 Hz in binary format, although a separate 1 Hz ASCII text file was also written. This text file incorporated data processed to 1 Hz from the AT4 output, and allowed the data to be looked at with a minimal amount of processing at the end of a flight. Once again, Excel and Igor Pro were used for data processing and analysis. Since the AIMMS data was logged independently, the datasets are synchronised by reference to the GPS satellite times recorded by the AIMMS and AT4, which should be consistent for the two instruments.

When analysing data as a function of position, an averaging period of 30 s was used (equivalent to a spatial resolution of around 1.7 km at a typical cruise speed of  $55 \text{ m s}^{-1}$ ), enabling large-scale features to be readily identified. This data was overlaid on terrain relief information taken from the US Geological Survey global 30 arc-second digital elevation model (GTOPO30), giving a spatial resolution of 1 km. The horizontal wind vector from the AIMMS was displayed along with data from ground-based instruments, including the UFAM wind profiler, and the University of Leeds AWS mesonet.

In terms of comparing air turbulence between different flights and flight legs, the variances of the horizontal and vertical wind components,  $\sigma_w^2$  and  $\sigma_{ff}^2$ , were of most use, along with the turbulent kinetic energy (TKE). Here,  $f$  is the magnitude of the

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horizontal wind calculated from the east ( $u$ ) and north ( $v$ ) components.

In order to determine whether the AIMMS was sampling all the scales of air motion, power spectral densities were computed for the wind components. When the power spectrum is plotted on log-log axes, a gradient of  $-5/3$  should be expected across a broad range of frequencies known as the inertial sub-range (Leslie, 1973), indicating that all scales of turbulence are being sampled adequately.

## 5.5 Performance and characterisation

It is always necessary when using a new instrument to determine the quality of the data, to ensure that the installation optimises the performance of the probe. A test flight from Liverpool airport and CSIP Flight 8 on 19 August (see Table 1) have been used to compare the position and attitude data with that obtained from the high precision Javad AT4. The ability to correctly map out a wind field was also examined, with Flight 8 being chosen as a day with a consistent wind flow across the CSIP area, allowing a comparison with ground-based instruments.

### 5.5.1 Comparison with the Javad AT4

A test flight was carried out from Liverpool airport in January 2006 which contained a number of tight manoeuvres under the direction of air traffic control. This offered an ideal opportunity to use the data to compare the performance of the AIMMS DGPS with the AT4.

It was assumed that the two systems reported position instantaneously since these measurements require minimal processing. On this basis, the AIMMS reports heading, pitch and roll up to 2 s after the AT4, as expected due to the use of the Kalman filter processing method in the AIMMS data (see Sect. 5.1). Taking these time lags into account, the signals for position (i.e. latitude and longitude), heading, pitch and roll were then compared. The regressions for these comparisons are shown in Table 3.

Agreement for position, heading and roll is clearly excellent. The pitch comparison

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is not as good, mainly due to discrepancies in magnitude. Further 5 Hz data from Flight 8 show that, unlike in straight and level flight, the time lag between the AIMMS and AT4 reduces in tight manoeuvres. This may reflect the capability of the AIMMS to measure attitude at 40 Hz, with the phase difference decreasing at higher rates of change of pitch: in effect the AT4 reacts less quickly to rapid changes in attitude than the AIMMS. The relatively large offsets in the pitch and roll regressions are due to the AT4 calibration using a different baseline to the AIMMS, having been calibrated with the aircraft stationary on the ground.

### 5.5.2 CSIP Flight 8 – uniform wind field

Flight 8, during IOP17 on 19 August, used the northern box route, flying the circuit twice (see Table 1). This flight was chosen because the conditions on the day showed a moderate and consistent wind speed across the CSIP flying area. Developing cloud heads led to prolonged rain over the eastern half of the CSIP area and flash floods near the south coast in the morning, delaying the start of the IOP. During the early afternoon scattered showers broke out in Wales and the north-west midlands, followed by the formation of a line of showers just west of Bath from 13:45 UTC. These showers reached peak intensity at 15:00 UTC with tops at 6 km, weakening between 16:00–17:00 UTC, by which time the showers extended from east of Bath to the Isle of Wight.

Figure 5 shows the wind speed and direction from the AIMMS as a function of position along the flight track. Wind direction was consistently NW-NNW across the whole of the flight area, whilst wind speed showed a gradient from  $10 \text{ ms}^{-1}$  in the NW to  $5 \text{ ms}^{-1}$  in the SE. Also shown in Fig. 5 is the output from the UFAM wind profiler located at Linkenholt. This instrument gives wind speed and direction up to a height of 3.5 km above ground. The data shown in Fig. 5 is averaged over the period of the southern leg of the flight at a height of 690 m a.m.s.l., this being the closest level to the Cessna flight, and shows good agreement with the AIMMS data.

Additionally, Fig. 5 includes data from the AWS mesonet, with each measurement being time-matched to the closest approach by the Cessna. Although this data is taken

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at 1 m above ground, it is clear there is good agreement with the airborne data from the Cessna with slight changes in the AIMMS wind direction across the measurement area being reflected well in the AWS data.

### 5.5.3 Power Spectral Density

5 For this flight, only 5 Hz data were available so power spectral density (PSD) information is only available up to 2.5 Hz. The ideal form for the PSDs is a uniform slope in the inertial sub-range of gradient  $-5/3$  on a log-log plot, indicating that the probe is sampling all relevant frequencies correctly. Figure 6 shows the PSDs for pitch, roll, vertical wind speed and horizontal wind speed. Whilst not completely uniform in slope, 10 the PSDs for pitch and roll and horizontal wind speed do not deviate significantly from the ideal case. However, for the vertical wind component there is a significant deviation between 0.6–1.0 Hz. The reason for this is not understood at present, and would require data at higher frequencies to allow a more complete analysis.

## 6 Assessment and errors

15 Comparison with the AT4 during tight aerial manoeuvres, and with ground-based instruments during CSIP Flight 8 show the AIMMS to be working satisfactorily. Of particular significance, the ability to map a wind field accurately is demonstrated. Given the good agreement with other instruments, the figure of  $\pm 0.5 \text{ ms}^{-1}$  for the accuracy of the horizontal wind speed given by Aventech is realistic. Power spectral densities conforming well to the ideal also give confidence in the horizontal wind speed data. Since 20 the wind direction is computed from the two horizontal wind components, the accuracy of the wind direction will clearly be a function of the magnitude of those components, although at low wind speed this parameter is naturally highly variable.

25 Accuracy for position is not in doubt: both the AIMMS and the AT4 use the differential method for calculating position which is accurate to within 0.01 m. Similarly, data for

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roll and heading agree extremely well between the two GPS systems. Data for pitch do not agree as closely, however it is thought the data from the AT4 suffers from variable lag compared to the AIMMS due to the manner in which it is calculated.

No estimate is made for errors in data computed in turns – for the purposes of this paper, the AIMMS was not calibrated to give reliable data under such conditions. There was also no means of determining the absolute accuracy of the vertical wind speed, since there was no comparable independent dataset, although the power spectra suggest that some of the variance in the vertical wind speed is being lost.

## 7 Case studies: Flights 6 and 7 during IOP16

### 7.1 Synoptic situation

In contrast to the case study of Flight 8 discussed in Sect. 5.5, wind conditions indicated by the AIMMS during Flights 6 and 7 showed considerable variation across the CSIP area, indicating a convergence zone in the centre of the Northern box. The CSIP area was covered by a weak, mostly southerly flow ahead of cold-frontal cloud belt advancing slowly from the west. This resulted in a warm sunny day over much of the region, with only shallow convection to start with. The first cumulus, at around 1000UTC were closely associated with topography in SE England and the Midlands. These soon developed over the whole area, but were restricted to a height of 3 km by a strong lid. By late afternoon the lid had weakened, and by 16:00 UTC cloud top had reached 6 km.

It is the presence of the convergence zone which makes this a critical test of the AIMMS, in that it requires the probe to be able to determine subtle variations in the wind field as well as more significant changes, often along a single flight leg. The ability to successfully map such a wind field would confirm the viability of this probe for atmospheric research.

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The wind speed and direction from the AIMMS as a function of position is shown in Fig. 7, which also shows data from the AWS mesonet and the UFAM wind profiler, with the Cessna flying the northern box route. As before, the AWS and profiler data are time-matched for the closest approach by the Cessna. Comparison with the wind profiler is discussed in Sect. 7.4.

The wind speed was light and variable along the northern legs, becoming more steady further south at  $5\text{--}6\text{ ms}^{-1}$ . The most remarkable feature of this flight is the dramatic change in wind direction from SW in the west to SE in the east which results in an apparent convergence on the centre of the northern box. As with the data for Flight 8, the AWS mesonet agrees well with the AIMMS in terms of the broad pattern of wind circulation, although the AWS data appears to show that the convergence occurs more towards the southern edge of the box. Whilst it must be remembered that the AWS is representative of conditions at the surface, it is extremely encouraging in terms of assessing the quality of the AIMMS data that the AWS mesonet confirms the existence of a convergence zone.

As expected, there were no significant trends in vertical wind speed, although there were some larger updrafts towards the eastern end of the southern leg of the box, where the standard deviation in the vertical wind speed was also high. Vertical wind speed was also highly variable on the eastern leg, although the average value was close to zero.

Power spectral densities computed for this flight were found to be very similar to those shown for Flight 8 (see Fig. 6). Given the nature of the converging wind field, PSDs were calculated for all legs, but were found to be consistent. In general, levels of turbulence, indicated by the variances of the horizontal and vertical wind components, were higher on the northern and eastern legs of the flight, corresponding with those areas experiencing greater mesoscale variability in wind direction and speed. Figure 8 shows values of  $\sigma_w$ ,  $\sigma_{ff}$ ,  $\sigma_T$ ,  $\sigma_{RH}$ , TKE,  $dd$ ,  $ff$  and  $w$  for Flights 6 and 7 for the purposes

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of comparison as a function of position on the flight track.

### 7.3 Flight 7, IOP16, 18 August, 14:09–15:51

This flight took place on the same day as Flight 6, but occurred further on into the convective development phase. In general the wind field was similar to Flight 6, but with slightly higher wind speeds. The northern box was flown twice, with consistent patterns in the wind field on the two passes.

Figure 9 shows wind speed and direction from the AIMMS as function of position for Flight 7, along with time-matched AWS and profiler data. Horizontal wind speed reached  $7\text{--}8\text{ ms}^{-1}$  in the west, but dropped off almost to zero in the NE corner of the box. Once again, the marked feature of this flight was the dramatic change in wind direction across the box, from westerly in the west to easterly in the east, and again an apparent convergence towards the centre of the northern box. The area with the lowest wind speeds appeared to have moved eastwards along the northern edge of the box by approximately 30 km over  $2\frac{1}{2}$  h between Flight 6 and the second pass of Flight 7. Once again, the broad pattern of circulation is confirmed by the ground-based AWS mesonet data and also by the point measurement provided by the wind profiler.

Vertical wind speed remained highly variable on the eastern leg of the box for both passes (Fig. 8). In general standard deviations of  $w$ ,  $u$ ,  $T$  and RH were relatively high for much of the northern and eastern legs of the flight. Combined with the variability in the wind direction in these areas, for both this and Flight 6, this strongly suggests that a broad-scale meteorological feature was moving from west to east across the upper part of the box during the course of Flights 6 and 7. This feature is likely to have been a convergence line or zone, particularly given the striking pattern of wind direction observed by the AIMMS.

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## 7.4 Comparison with the wind profiler for Flights 6, 7 and 8

The UFAM wind profiler operated by the University of Manchester was the only ground-based instrument to measure the wind vector close to the flight path of the Cessna during these flights. The profiler is a three antenna Doppler radar (UHF PCL1300) manufactured by Degreave Horizon. It is designed to measure the three components of the wind vector twenty four hours a day to an accuracy of less than  $1 \text{ ms}^{-1}$  for speed and less than  $10^\circ$  for direction. The frequency of operation was 1290 MHz (L-band) with a peak power of 3.5 kW and a beam width of  $8^\circ$ . The vertical range was between approximately 70 m and 3940 m with a resolution of 72 m. It must, however, be remembered that the profiler did not lie exactly on the flight path at any point, and that any comparison with the AIMMS data is therefore complicated by this lateral displacement. A full description of the profiler is given by Norton et al. (2006).

Table 4 shows a comparison of wind direction and horizontal and vertical wind speed from the AIMMS and the wind profiler, with the errors reflecting the spread of 1-min averaged values used to calculate the figures. Data is only shown for those legs of the flights which pass closest to the profiler. Qualitative comparison between the two datasets is generally good for horizontal wind speed. Agreement is not quite so good for wind direction, although for both speed and direction agreement is quantitatively very good for Flight 8. This reflects the general stability of conditions experienced across the whole of the CSIP area for Flight 8. For Flights 6 and 7 the wind profiler in particular shows how variable conditions were on even a short timescale, with wind direction changing markedly with height as well as time. The effect of this variability is exacerbated by the lateral separation of the measurements, making it difficult to compare vertical wind speeds, although differences between the two instruments are not significant.

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## 8 Conclusions

A low-cost turbulence probe has been installed and used on the UFAM Cessna 182J, with encouraging results. The data has shown good comparison with other measurement methods, including ground-based observations and ground-based remote sensing. In particular, the probe has proved to be capable of reliably mapping out a horizontal wind field over a large area.

The full capabilities of the probe in respect of assessing the absolute frequency response have not been tested for technical reasons, but all the available data shows the probe to be functioning in a manner consistent with being used for scientific research. Whilst the frequency response for horizontal wind speed and attitude is good, further investigation is required with higher frequency data to test the full response of the probe for vertical wind speed.

Good quality data from this probe is entirely reliant on successful calibrations, but the procedures for these ensure the probe is easily installed and operated on most types of aircraft. The recent addition of a second CPM to the system has made the AIMMS more useful as an on-line tool, as well as still being able to record full high frequency data: this is a significant deficiency in the standard system.

Further work will be carried out with 20 Hz data from the AIMMS. It is anticipated, from the data presented above, that the probe will in future be used to measure fluxes of atmospheric constituents such as trace gases and particulates.

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**Table 1.** Cessna flights during the main CSIP campaign in 2005.

Date	Time	Flight#	IOP	Flight Plan and way-points
29 June	10:43–11:40	1	5	East-West, 1 circuit, ABB'JKJA
4 July	10:48–11:35	2	6	East-West, 1 circuit, AJKI'A
13 July	10:23–13:12	3	8	East-West, partial Northern Box, 2 circuits, ABB'CIAJKABB'CIAJKI'A
13 July	14:38–16:23	4	8	East-West, partial Northern Box, 1 circuit, AJCIAJKAJCIA
11 August	15:46–16:57	5	14R	Southern Box, 1 circuit, ABLMNOPBA
18 August	12:18–13:08	6	16	Northern Box, 1 circuit, AB'CDEFI'A
18 August	14:10–15:51	7	16	Northern Box, 2 circuits, AB'CDEFJCDEFB'A
19 August	14:11–16:14	8	17	Northern Box, 2 circuits, ABJCDEFGHDEFJA
25 August	08:05–10:11	9	18	Northern Box, 1 circuit, ABB'CDEFGHIA
25 August	13:24–14:44	10	18	North-South, 3 repeats, ABIBIBIA

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**Table 2.** AIMMS specifications.

Module	Specification
All modules	power: 12.5-37VDC, 800-900mA at 12.5V digital serial communication via 115kbaud controller area network Measurement frequency up to 20Hz
ADP	Overall weight 3.36 kg Operating range $-20^{\circ}\text{C}$ to $+50^{\circ}\text{C}$ Static pressure range 0–110 kPa, accuracy 0.1 kPa $\pm$ 0.05% Pitot pressure range 0–14 kPa, accuracy 0.02 kPa $\pm$ 0.05% Wind speed horizontal: accuracy 0.5 ms <sup>-1</sup> vertical: accuracy 0.75 ms <sup>-1</sup> Temperature glass encapsulated bead thermistor calibrated accuracy 0.05 $^{\circ}\text{C}$ (0.3 $^{\circ}\text{C}$ with dynamic heating correction) resolution 0.01 $^{\circ}\text{C}$ , time constant <5 s mounted in ventilated reverse-flow chamber Relative Humidity thermoset polymer capacitive accuracy $\pm$ 2% RH (0–100%) resolution 0.1%, time constant <5 s at 20 $^{\circ}\text{C}$ mounted in ventilated reverse-flow chamber
IMU	Compass heading 3-axis magnetic field sensors Communication external RS232 at 115kbaud Overall weight 0.74 kg Accelerometer 3-axis, range $\pm$ 5 g, accuracy 0.005 g, operated at 40 Hz
GPS	Rate gyro 3-axis, operated at 40 Hz Overall weight 0.80 kg Heading accuracy 0.1 $^{\circ}$ Position accuracy 0.01 m Antennae shares two dual frequency antennae with AT4 DGPS
CPM	Overall weight 0.60 kg Processor Motorola DSP56F807 processor 16Mbit flash memory external RS232 at 115kbaud

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**Table 3.** Comparison between AIMMS and Javad AT4 attitude and position data.

Measurement	Regression	Regression	R <sup>2</sup>
	Slope ( $\pm$ std. err)	Intercept ( $\pm$ std. err)	
Latitude	0.999 ( $\pm$ 0.000)	+0.030 ( $\pm$ 0.002)	1.00
Longitude	1.000 ( $\pm$ 0.001)	+0.000( $\pm$ 0.0003)	1.00
Heading	1.000 ( $\pm$ 0.000)	+0.300( $\pm$ 0.006)	0.99
Pitch	0.874 ( $\pm$ 0.004)	−3.078 ( $\pm$ 0.003)	0.62
Roll	0.999 ( $\pm$ 0.001)	+1.912( $\pm$ 0.003)	0.99

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**Table 4.** Comparison between the radar wind profiler and the AIMMS.

Measurement		Flight 6 profiler	AIMMS	Flight 7 profiler	AIMMS	Flight 8 profiler	AIMMS
horizontal wind speed $\text{ms}^{-1}$	pass 1	$1.7 \pm 0.4$	$5.1 \pm 0.5$	$2.3 \pm 1.5$	$4.6 \pm 0.8$	$3.8 \pm 2.0$	$5.6 \pm 0.4$
	pass 2	$2.7 \pm 1.5$	$5.0 \pm 0.6$	$4.4 \pm 2.0$	$3.6 \pm 0.7$	$4.9 \pm 1.5$	$5.3 \pm 1.2$
	pass 3	–	–	$2.3 \pm 2.0$	$3.3 \pm 0.6$	$5.9 \pm 2.4$	$6.3 \pm 0.8$
wind direction degrees	pass 1	$180 \pm 73$	$143 \pm 7$	$280 \pm 64$	$139 \pm 10$	$341 \pm 36$	$311 \pm 21$
	pass 2	$138 \pm 113$	$175 \pm 4$	$172 \pm 112$	$193 \pm 18$	$315 \pm 17$	$307 \pm 10$
	pass 3	–	–	$344 \pm 50$	$207 \pm 20$	$314 \pm 14$	$308 \pm 6$
vertical wind speed $\text{ms}^{-1}$	pass 1	$-0.09 \pm 0.41$	$0.14 \pm 0.47$	$0.56 \pm 0.45$	$0.30 \pm 0.21$	$0.01 \pm 0.01$	$0.03 \pm 0.05$
	pass 2	$-0.75 \pm 0.60$	$0.13 \pm 0.47$	$0.39 \pm 0.51$	$-0.02 \pm 0.21$	$0.00 \pm 0.02$	$0.12 \pm 0.12$
	pass 3	–	–	$0.10 \pm 0.50$	$0.03 \pm 0.14$	$0.01 \pm 0.02$	$0.03 \pm 0.01$

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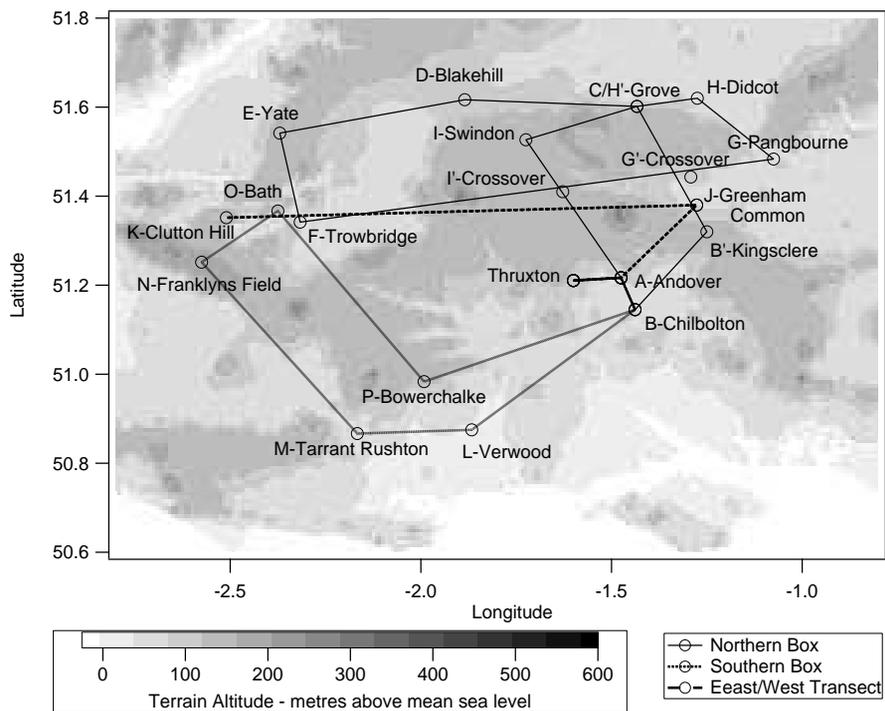
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## Turbulence measurements using the AVENTECH AIMMS20AQ

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**Fig. 1.** Flight paths and way-points for the Cessna, showing the pre-determined northern box, southern box and east/west transects.

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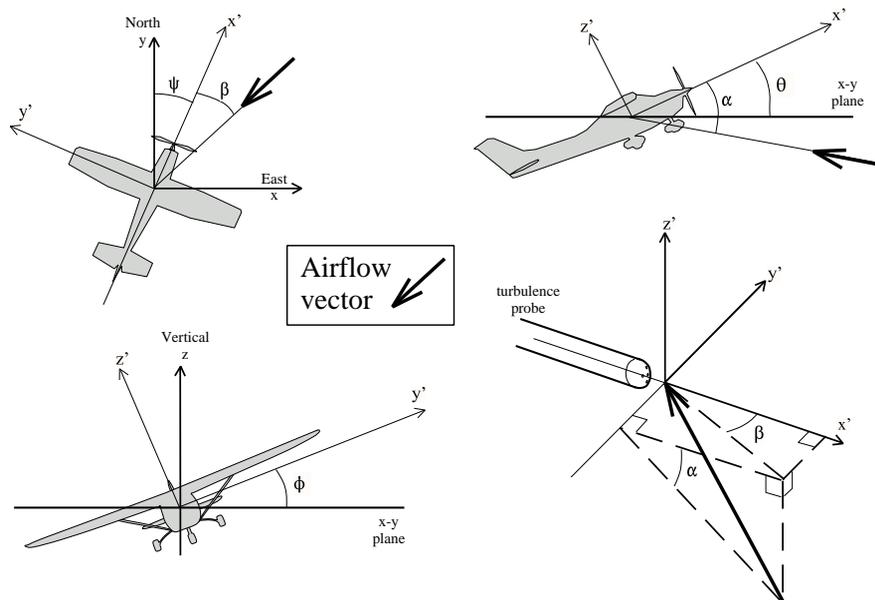
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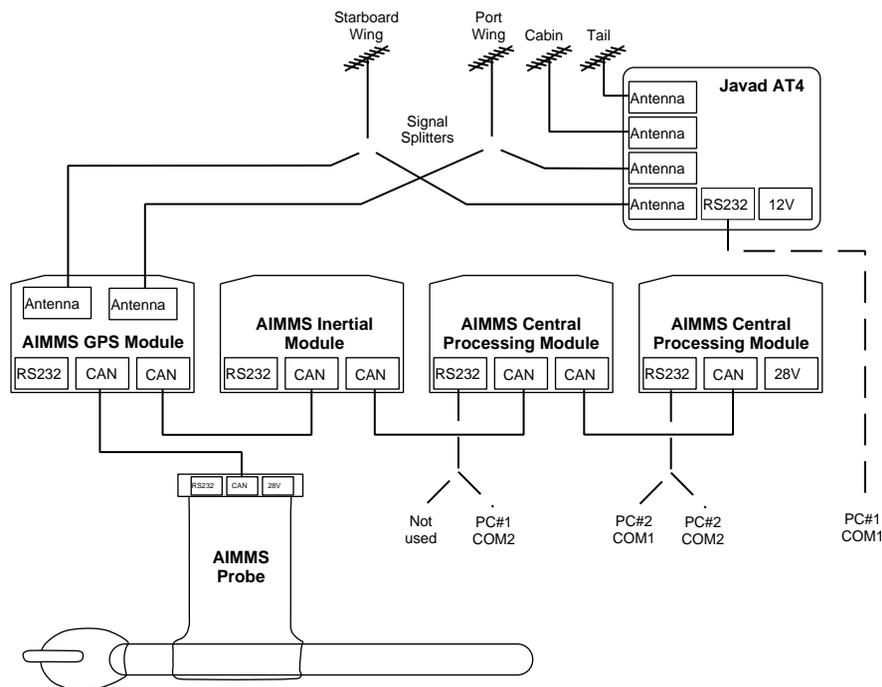


**Fig. 2.** The relationship between the Earth and aircraft coordinates. Earth coordinates are North, East and vertical, or  $xyz$ . Aircraft coordinates are longitudinal axis, lateral axis and vertical axis, or  $x'y'z'$ .  $\Psi$  = heading in  $xyz$ ;  $\theta$  = pitch in  $xyz$ ;  $\phi$  = roll in  $xyz$ ;  $\alpha$  = angle of attack in  $x'y'z'$ ;  $\beta$  = sideslip angle in  $x'y'z'$ .

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**Fig. 3.** Schematic of the Javad AT4 differential GPS and AIMMS modules. The figure shows the optional second CPM unit which allows on-line data to be displayed whilst the first CPM records high frequency data.

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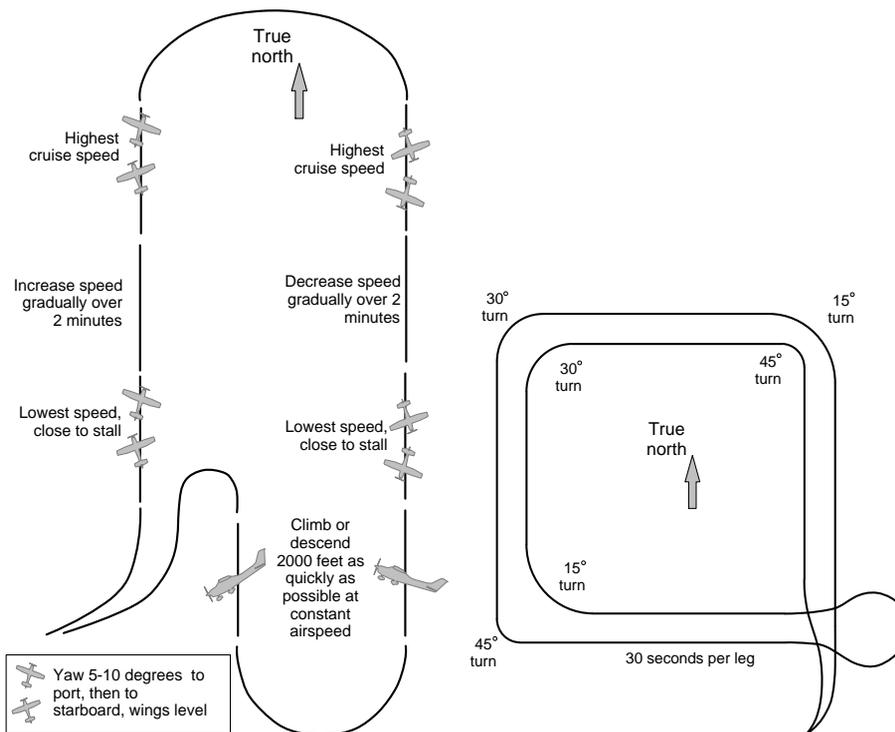
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## Turbulence measurements using the Aventura AIMMS20AQ

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**Fig. 4.** Schematic diagram of the two airborne calibration procedures for the AIMMS sensor: **(a)** shows the flight plan for the aerodynamic calibration, whilst **(b)** shows the flight plan for the cross-axis calibration.

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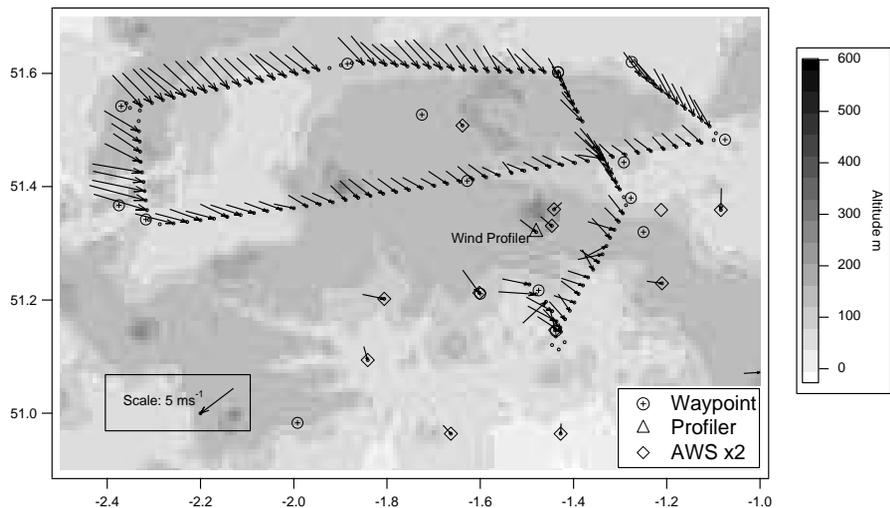
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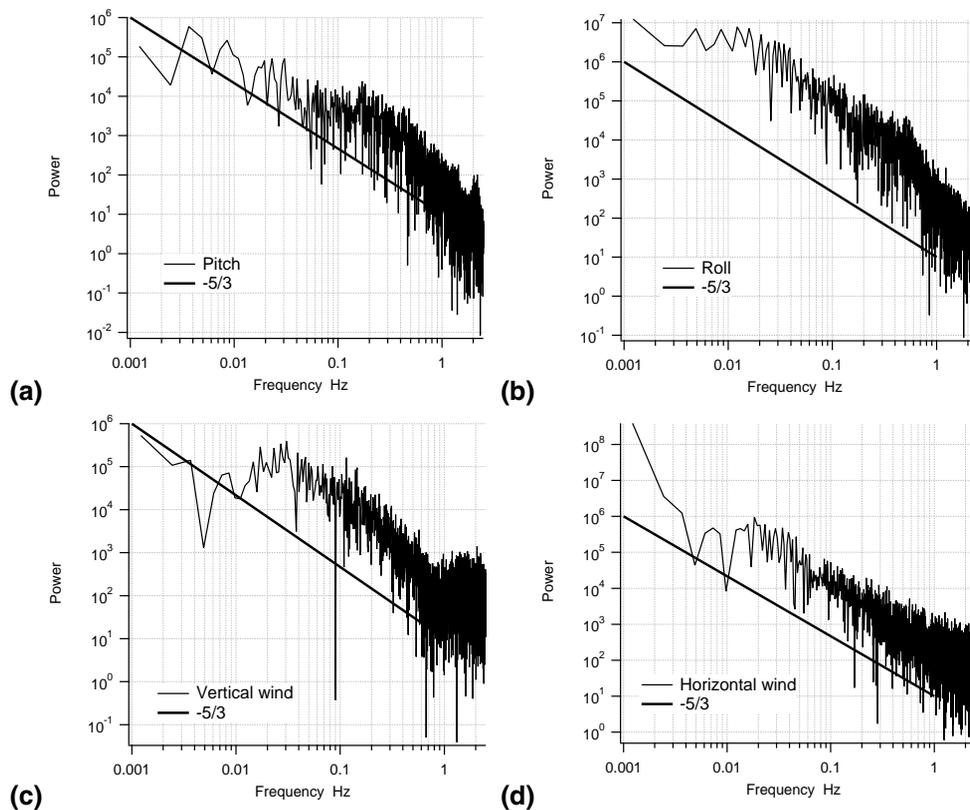


**Fig. 5.** Horizontal wind vector for Flight 8, including data from the wind profiler and the AWS mesonet. For clarity, only the first pass round the northern box is shown, and the AWS data is scaled by a factor of 2.

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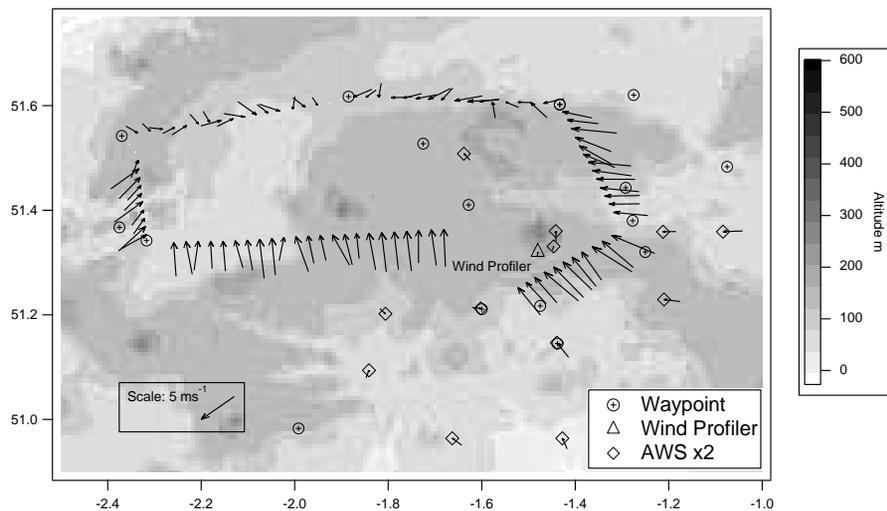


**Fig. 6.** Power spectral densities for AIMMS data from Flight 8 for **(a)** pitch, **(b)** roll, **(c)** vertical wind and **(d)** horizontal wind.

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**Fig. 7.** Horizontal wind vector for Flight 6, including data from the wind profiler and the AWS mesonet. The AWS data is scaled by a factor of 2.

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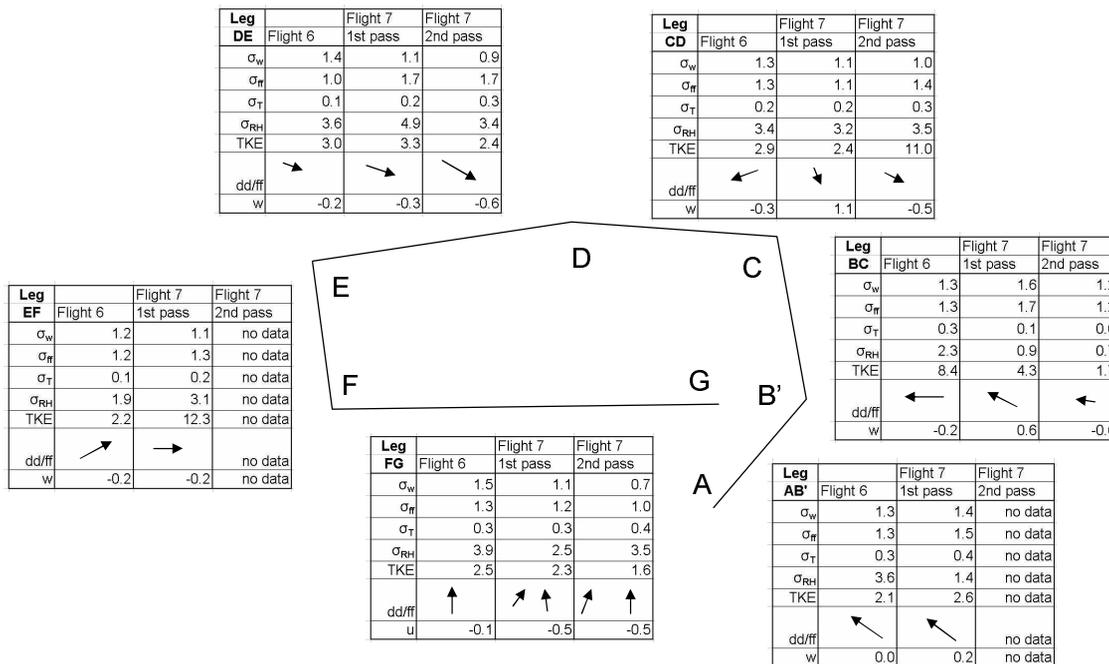


Fig. 8. Summary of standard deviations and means for Flights 6 and 7, to show relative variations as a function of flight path.  $\sigma_w$ ,  $\sigma_{ff}$  and  $u$  are in  $\text{ms}^{-1}$ ,  $\sigma_T$  is given in  $^{\circ}\text{C}$ ,  $\sigma_{RH}$  is given in %, and TKE is given in  $\text{m}^2 \text{s}^{-2}$ . dd/ff shows qualitatively the wind direction and strength.

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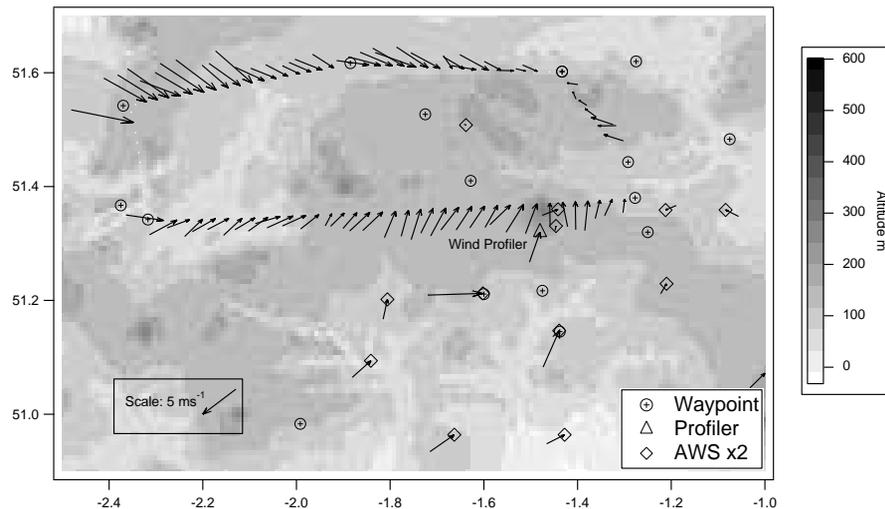
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**Fig. 9.** Horizontal wind vector for Flight 7, including data from the wind profiler and the AWS mesonet. For clarity, only the first pass round the northern box is shown, and the AWS data is scaled by a factor of 2.

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